

1. Record Nr.	UNINA9910716562003321
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Titolo	Analytical and experimental studies of shock interference heating in hypersonic flows // by J. Wayne Keyes and Frank D. Hains
Pubbl/distr/stampa	Washington, D.C. : , : National Aeronautics and Space Administration, , May 1973
Descrizione fisica	1 online resource (iv, 137 pages) : illustrations
Collana	NASA/TN ; ; D-7139
Soggetti	Fluid mechanics Shock waves Aerodynamic heating Aerodynamics, Hypersonic Boundary layer
Lingua di pubblicazione	Inglese
Formato	Materiale a stampa
Livello bibliografico	Monografia
Note generali	"May 1973."
Nota di bibliografia	Includes bibliographical references (pages 34-39).
Sommario/riassunto	An analytical and experimental study is presented of the aerodynamic heating resulting from six types of shock interference patterns encountered in high speed flow. Centerline measurements of pressure and heat transfer distributions on basic bodies were obtained in four wind tunnels for Mach numbers from 6 to 20, specific heat ratios from 1.27 to 1.67, and free stream Reynolds numbers from 3 million to 25.6 million per meter. Peak heating and peak pressures up to 17 and 7.5 times stagnation values, respectively, were measured. In general, results obtained from semiempirical methods developed for each of the six types of interference agreed with the experimental peaks.